

Faculty of Engineering

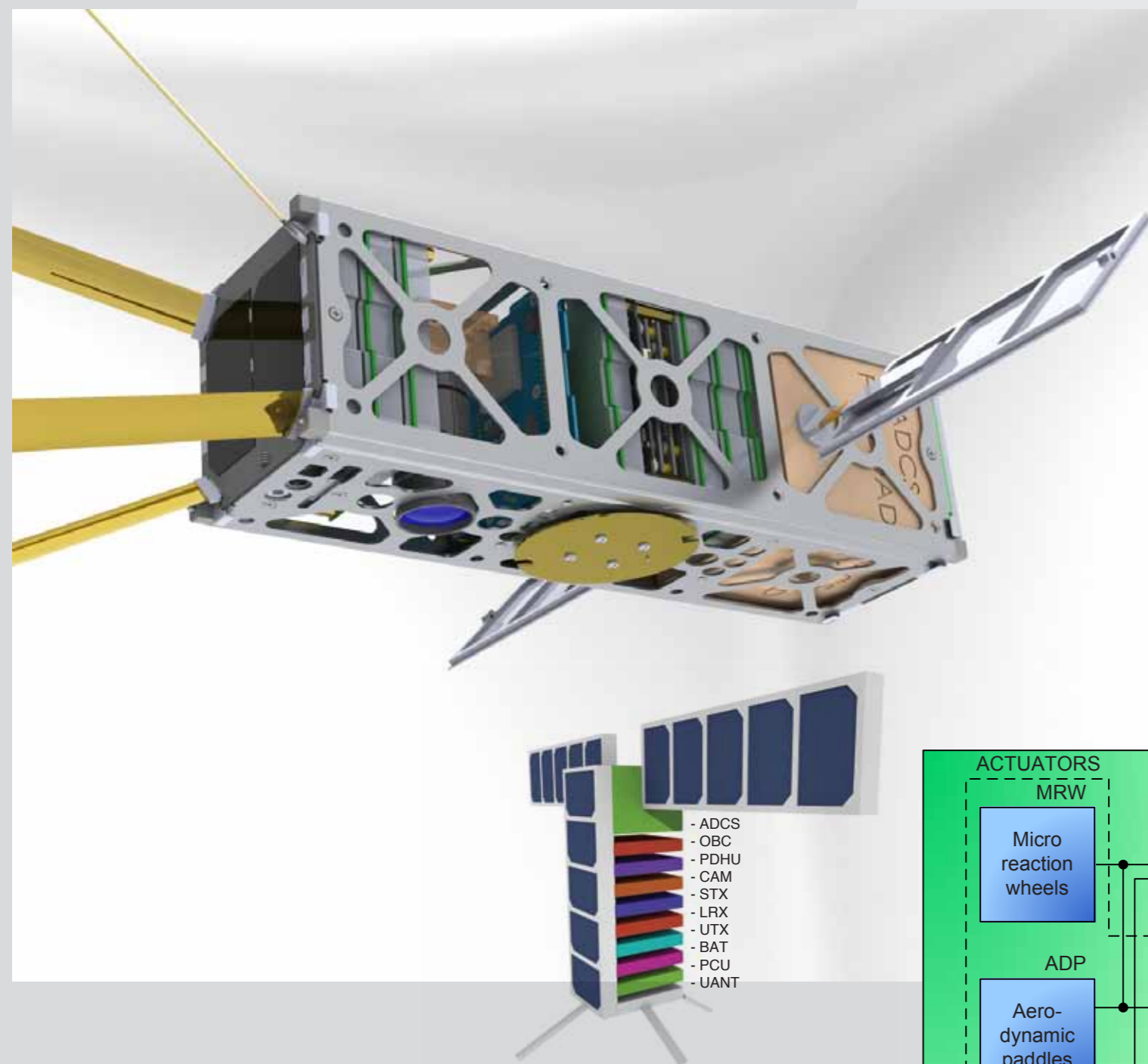
# CubeSats: Development Outline

## 1ST MISSION

### 1. CubeSat

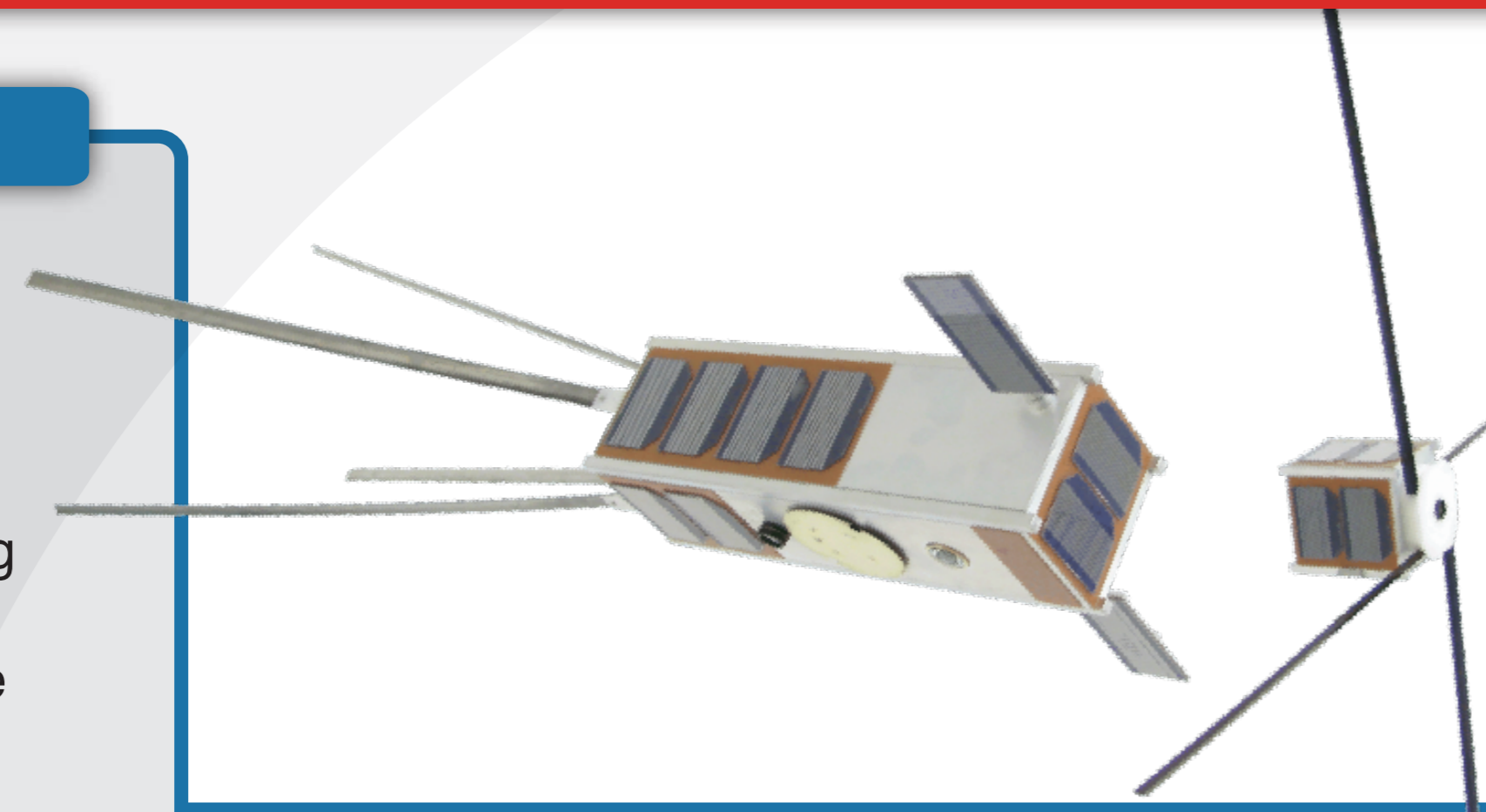
- 3U CubeSat: 10 x 10 x 30 cm
- Based on an international standard, making space more affordable.
- Subsystem use PC104 standard, stackable PCBs

### 2. Concept design



### 3. 3U CubeSat payloads

- Omnivision 5MP Colour Matrix Camera, Parameters: Aperture  $\approx$  20mm, Focal length  $\approx$  50mm, Ground resolution  $<$  50m, Swath width  $\approx$  130km
- L-band to S-band transponder
- Store & forward system on UHF, AX.25 protocol



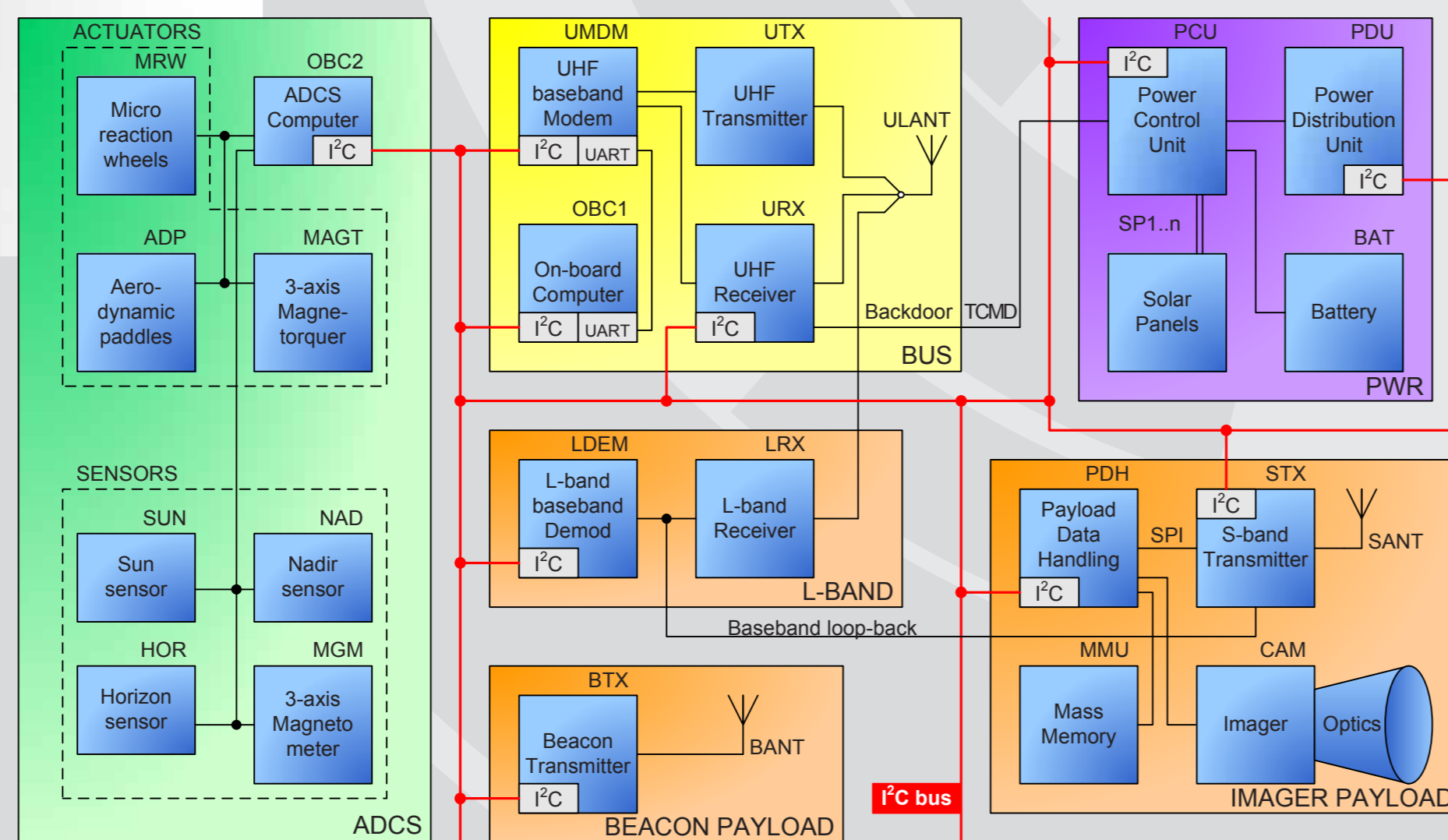
### Major milestones for satellites:

- Integrated 3U prototype satellite by Oct 2011
- 1U Flight model expected Aug 2011 with possible launch opportunity early 2012
- Prototypes of both satellites to be showcased at IAC 2011

### 4. Subsystems

- UHF TT&C transceiver
- S-band payload transmitter
- L-band receiver
- Full 3-axis stabilisation ADCS by University of Stellenbosch, deployed aerodynamic paddles and passively stabilised "tail feathers". Sensors: Nadir, sun, horizon and magnetometer. Control: Actuators, magnetorquers, nano reaction wheels.
- OBC based on Pumpkin OBC featuring MSP430

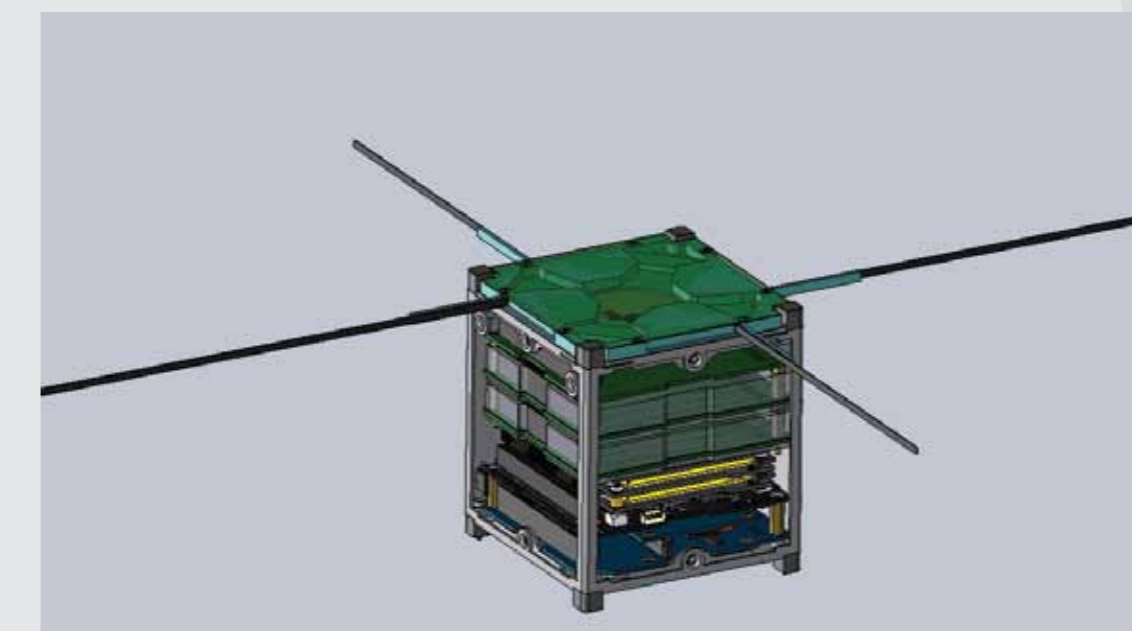
### 5. Block diagram



## 2nd MISSION (FAST-TRACKED)

### 1. CubeSat

- Fast track satellite
- 1U CubeSat: 10 x 10 x 10 cm
- Science payload in collaboration with SANSA Space Science



### 2. 1U CubeSat payloads & subsystems

- VGA Colour Matrix Camera with onboard EEPROM for image storage.
- 3-axis stabilised ADCS with torquer rods.
- HF beacon for calibration of space weather radar antennas at SANAE, featuring a novel HF antenna deployment mechanism
- VHF/UHF TT&C communication link
  - 9600 baud GMSK, 1200 baud AFSK modems
  - DTMF backdoor
- Power system: Clyde Space batteries
  - Lithium-polymer 4.1V, 1.25Ah
  - Clyde Space solar panels
  - 2.2V per cell at MPP, 1W max
  - 28% efficiency triple junction cells

